

Orbit in Space

- $x, y =$ north, west on sky plane thru m_1 at $x,y=0$, observer at $-z$.
- $i =$ inclination between sky and orbit planes
- $N =$ ascending node, m_2 crosses $x-y$ plane in $+z$ direction
- $\varpi =$ longitude of ascending node
- $\omega =$ longitude of periastron
- $\theta =$ true anomaly

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Relative orbit on the Sky

$$x = x' \cos \Omega + y' \sin \Omega \cos i$$

$$y = x' \sin \Omega - y' \cos \Omega \cos i$$

$$z = y' \sin i$$

$$x' = r \cos(\mathbf{q} + \mathbf{w})$$

$$y' = r \sin(\mathbf{q} + \mathbf{w})$$

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Orbital Elements

specific angular momentum :

$$L_x = L \sin i \sin \Omega$$

$$L_y = -L \sin i \cos \Omega$$

$$L_z = -L \cos i$$

7 orbital elements :

size (a), shape (e),
 orientation in space (i, \mathbf{w}, Ω)
 and in time (T, P).

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Spectroscopic Binaries

- m_2 at $(r, \mathbf{q} + \mathbf{w})$
 - project along the line of nodes: $r \cos(\mathbf{q} + \mathbf{w})$
 - perpendicular to line of nodes: $r \sin(\mathbf{q} + \mathbf{w})$
 - project along line of sight: $z = r \sin(\mathbf{q} + \mathbf{w}) \sin i$
- radial velocities along line-of sight

$$V_{rad} = \dot{z} = \sin i \left[\dot{r} \sin(\mathbf{q} + \mathbf{w}) + r \dot{\mathbf{q}} \cos(\mathbf{q} + \mathbf{w}) \right]$$

use $r = \frac{a(1-e^2)}{1+e \cos \mathbf{q}} \rightarrow \dot{r} = \frac{e \sin(\mathbf{q}) r \dot{\mathbf{q}}}{1+e \cos \mathbf{q}}$

and Kepler's 2nd Law $r^2 \dot{\mathbf{q}} = \frac{2pa^2(1-e^2)^{3/2}}{P}$

$$V_{rad} = \frac{2pa \sin i}{P \sqrt{1-e^2}} [\cos(\mathbf{q} + \mathbf{w}) + e \cos(\mathbf{w})]$$

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Spectroscopic Orbital Velocities

$$V_{rad} = K [\cos(\mathbf{q} + \mathbf{w}) + e \cos \mathbf{w}] + g$$

$$K = \frac{2pa \sin i}{P \sqrt{1-e^2}}$$

ascending node: $\cos(\mathbf{q} + \mathbf{w}) = +1$

$$V_{max} = K [e \cos \mathbf{w} + 1] + g$$

descending node: $\cos(\mathbf{q} + \mathbf{w}) = -1$

$$V_{min} = K [e \cos \mathbf{w} - 1] + g$$

semi-amplitude:

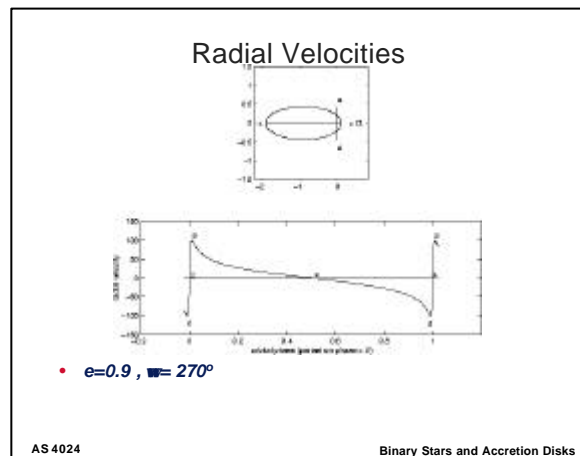
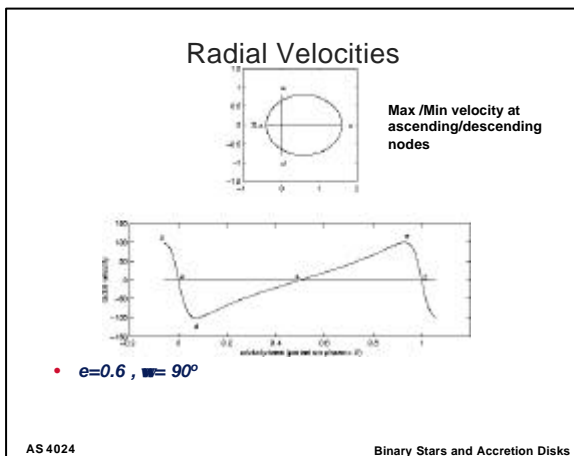
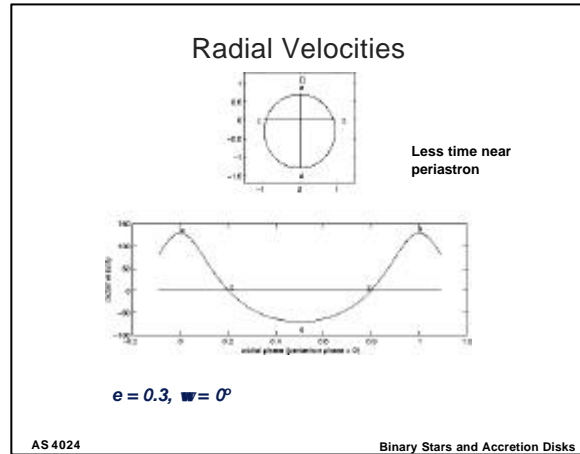
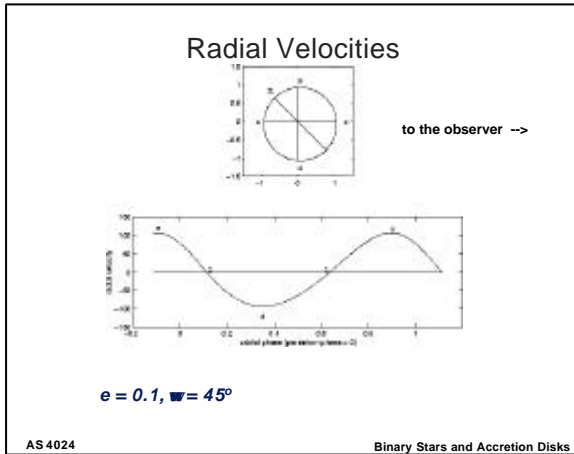
$$K \equiv (V_{max} - V_{min}) / 2$$

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Circular Orbit

- $K_1 = 100 \text{ km/s}$ $K_2 = 200 \text{ km/s}$ $q = m_2 / m_1 = 0.5$

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Orbits from radial velocities

- **Observations** n radial velocity measurements:
 - $V_r(i)$ at times $t(i)$ $i = 1 \dots n$
- **Elliptical orbit model:**

$$E - e \sin E = \frac{2\pi}{P}(t - T)$$

$$\tan\left(\frac{q}{2}\right) = \left(\frac{1+e}{1-e}\right)^{1/2} \tan\left(\frac{E}{2}\right)$$

$$V_r = K(\cos(q + \omega) + e \cos \omega) + I$$
- **Best fit determines** (K, e, ω, T, P)
- e.g. using a least squares procedure

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Minimum masses

SB2 : measure P, K_1, K_2, e

$$K_i = \frac{2\pi a_i \sin i}{P\sqrt{1-e^2}} \rightarrow a_i \sin i = \frac{\sqrt{1-e^2}}{2\pi} K_i P$$

$$m_1 = M \frac{a_2}{a} = \frac{4\pi^2 a^2 a_2}{G P^2}$$

$$m_1 \sin^3 i = \frac{4\pi^2 a^2 a_2 \sin^3 i}{G P^2} = \frac{(1-e^2)^{3/2} (K_1 + K_2)^2 K_2 P}{2\pi G}$$

SB1 : measure P, K_1, e , calculate the mass function :

$$f(m_2) = \frac{m_2^3 \sin^3 i}{M^2} = \frac{(1-e^2)^{3/2} K_1^3 P}{2\pi G}$$

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